TO: ALL HOLDERS OF WING L.E. FLAP ASSEMBLIES NO. 1 AND 4 OVERHAUL MANUAL, 57-56-31

REVISION NO. 8, DATED JUL 1/06

HIGHLIGHTS

DESCRIPTION OF CHANGE	TOPICS AFFECTED												
	D&O	D/Assy	Cleaning	Insp/Chk	Repair	Assy	F/C	Test	T/Shooting	S/Tools	Storage	IPL	L/Overhaul
Updated IPL Fig. 1104 to add additional nuts (1, 17), washers (2, 18) and bolts (3, 19)												X	

WING LEADING EDGE FLAP ASSEMBLIES NO. 1 AND NO. 4

57-56-31

BOEING P/N 65-46424-1, -2, -5 thru -10, -13, -14, -19 thru -30

AIRLINE P/N

THE FOLLOWING DIRECTIVES APPLY TO THIS SUBJECT:

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
		PRR 30117	Feb 15/69
		PRR 30720	Feb 15/69
		PRR 31232	Feb 15/69
57-1028, Rev 2		PRR 31310	Dec 10/70
		PRR 31960-1	Dec 10/71
		PRR 31960-10	Dec 10/71
57-1051		MC 3400-26K	Dec 10/71
		MR 39055-1	Dec 10/71
		PRR 32070-15	Jun 10/72
		PRR 32121-30	Dec 25/72
		PRR 32070-16	Jun 25/73
		PRR 32253	Jun 25/73
		PRR 32259	Jun 25/73
57-1077			Dec 25/73
		PRR 32757-3	Jul 5/79
		PRR 32494-7	Jul 5/79
57-1114			Jul 5/80
		PRR 33191	Dec 5/83
57-1215		MC 0310MK3135	Jun 5/91

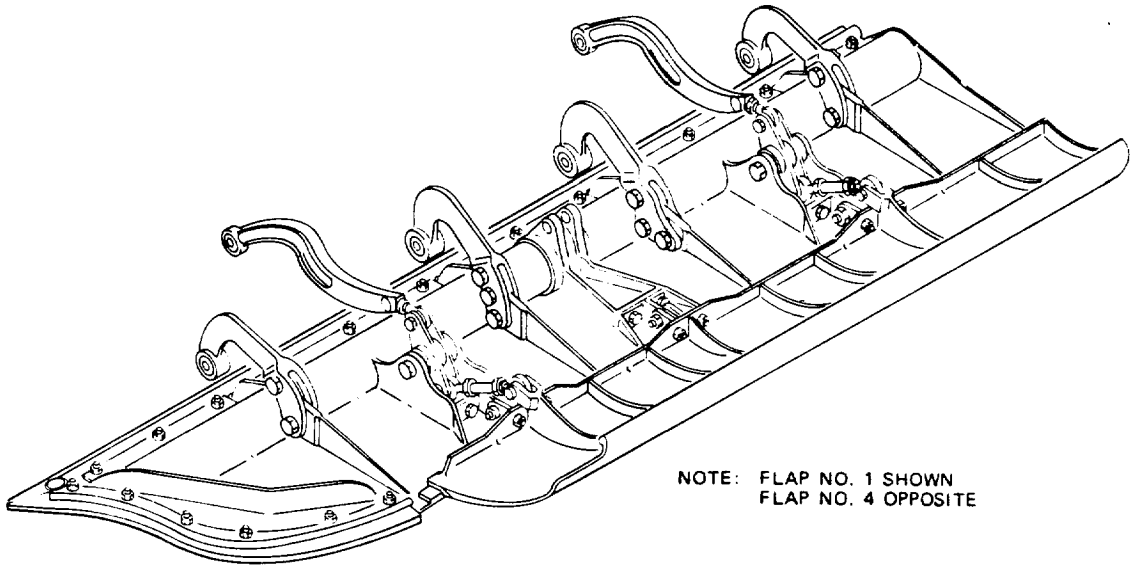
LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision
F Indicates foldout pages - print one side only

PAGE	DATE
57-56-31	
T-1	Jun 5/91
T-2	BLANK
* LEP-1	Jul 1/06
LEP-2	BLANK
T/C-1	Dec 10/71
T/C-2	BLANK
1	Jun 25/73
2	Feb 15/69
101	Jun 10/72
102	Jun 10/72
103	Jun 5/91
104	BLANK
201	Feb 15/69
202	BLANK
301	Jun 5/91
302	BLANK
401	Sep 1/94
402	Sep 1/94
403	Sep 1/94
404	Sep 1/94
501	Jun 5/91
502	Dec 25/72
503	Mar 1/95
504	Jan 5/81
505	Jan 5/81
506	BLANK
601	Dec 25/72
602	Dec 25/72
603	Jun 5/91
604	BLANK
901	Feb 15/69
902	BLANK
1101	Jul 5/79
1102	Jul 5/79
1103	Jul 5/79
1104	Jul 5/79
1105	Jul 5/79
1106	Jul 5/79
1107	BLANK
1108	Jan 5/81
1109	Jul 5/79
1110	Jan 5/81
1111	BLANK
1112	Jul 5/79
1113	Jul 5/79
1114	Jul 5/79
* 1115	Jul 1/06
* 1116	Jul 1/06
* 1117	Jul 1/06
* 1118	Jul 1/06
* 1119	Jul 1/06
* 1120	Jul 1/06

TABLE OF CONTENTS

WING LEADING EDGE FLAP ASSEMBLIES NO. 1 AND NO. 4



Wing Leading Edge Flap Assemblies No. 1 and No. 4
Figure 1

DESCRIPTION AND OPERATION

1. Description

   A. The lift capability of the 737 wing is supplemented by a set of three slats, and two Krueger type flaps, which are installed on the leading edge of each wing. These lightweight leading edge devices work in coordination with the trailing edge flaps to improve operation of the airplane at low speeds, and enable takeoff in minimum distance.

   B. Flap No. 1 is installed on the leading edge of the left wing, just inside the engine nacelle; flap No. 4 occupies an identical position on the right wing. The flaps are identical, except that they are opposites.

   C. Each flap is basically an aluminum alloy casting, with integral stiffeners and ribs. It is reinforced at attach points with steel bushings. A trailing edge, or tailgate structure, also of aluminum alloy, is attached to the aft edge to fair the assembly in with the wing structure. The entire assembly is attached to the front spar of the wing with four steel, gooseneck hinge fittings. The main flap holds a linkage assembly which moves the tailgate into position when the flap is operated.

2. Operation

A. Each leading edge flap is actuated by a two-position, cylinder type, hydraulic actuator. An integral part of each actuator is a locking device which locks the flap in either the fully extended or the fully closed position. This precludes the possibility of the flap being blown open or closed if hydraulic power fails.

DISASSEMBLY

1. Disassembly

NOTE: Flap assemblies, hinge assemblies and tailgate and lug assemblies are matched sets. Carefully note location of individual parts during disassembly to assure correct reassembly.

A. Removal of Linkage (See figure 1102.)

(1) Remove cotter pins (1), nuts (2), washers (3), and bolts (4). Remove link assemblies (5).

(2) Loosen checknut (7), and remove bearing (8), washer (9), and cylindrical nut (10) from each link (6). Remove checknuts (7).

NOTE: Do not remove bearing (11) from link (6) unless replacement is necessary.

(3) Remove cotter pins (12), nuts (13), washers (14), and bolts (15).

(4) Remove cotter pins (16), nuts (17), washers (18), bolts (19), and clamp-up bushings (20). Remove rod assemblies (21).

(5) Rod assembly (21, 69-61262-3):

(a) Remove lockwire, if applicable, loosen checknuts (23 and 23A) and remove rod end bearings (24 and 24A) from rod (22A). Remove checknuts (23 and 23A).

CAUTION: ROD (22A), CHECKNUT (23A), AND ROD END BEARING (24A) HAVE LEFT-HAND THREADS.

(6) Rod assembly (21, 69-37244-3):

(a) Remove lockwire, if applicable, loosen checknuts (23) and remove rod end bearings (24) from rod (22). Remove checknuts (23).

(7) Remove cotter pins (25), nuts (26), washers (27), and bolts (28). Remove crank assemblies (29).

NOTE: Do not remove bearings (31) or bushings (32) from crank assemblies (29) unless replacement is necessary.

B. Removal of Tailgate and Lug Assemblies (See figure 1103.)

(1) Remove items (11, 12, 13, and 14) and tailgate assembly (5) together with lug assemblies (15 and 18), and seal (32).

(2) Remove items (1 through 4) to remove lug assemblies (15 and 18) and shims (10).

NOTE: Note respective location of each lug assembly and attending shim to aid in reassembly. If shims are bonded in place, do not remove unless replacement is necessary.

Do not remove bushings unless replacement is necessary.

(3) Remove items (21 through 24) to remove lug assemblies (25) and shims (29).

NOTE: Note respective location of each lug assembly and attending shim to aid in reassembly. If shims are bonded in place, do not remove unless replacement is necessary.

Do not remove bushings unless replacement is necessary.

(4) Remove items (29 through 31) to remove seal (32) from tailgate (6).

NOTE: Do not remove bushings (7, 8 and 9) from tailgate (6) unless replacement is necessary.

C. Removal of Hinges (See figure 1104.)

NOTE: Note respective location of each hinge assembly and attending shims, and their relation to the flap, to aid in reassembly. If shims are bonded in place, do not remove unless replacement is necessary.

Do not remove bushings unless replacement is necessary.

(1) Remove nuts (1), washers (2), and bolts (3). Remove hinge assemblies (4) and shims (7). Remove sensor actuator (8).

(2) Remove nuts (9), washers (10), and bolts (11). Remove hinge assembly (12) and shims (15 and 16).

(3) Remove nuts (17), washers (18), and bolts (19). Remove hinge assembly (20) and shims (23 and 24).

D. Removal of Aerodynamic Seals (Fig. 1101)

(1) Remove nuts (1) and bolts (2). Remove retainer (3) and seals (4 and 5).

(2) Remove nuts (6) and bolts (7). Remove retainers (8, 9, 10, and 11) and seals (12 and 13).

(3) Remove rub strip (19) from basic flap (15) by scraping off with a sharp edged wooden or plastic tool.

NOTE: Do not remove bushings (16, 17, and 18) from flap (15) unless replacement is necessary.

CLEANING

1. General

   A. Wash and rinse all metal parts, except antifriction bearings, in cleaning solvent, Specification P-D-680, or equivalent.

   B. Remove stubborn accumulations of dirt with a stiff-bristle brush. Do not use a metallic brush.

   C. Dry all parts thoroughly with clean, lint-free cloth, or with dry. clean, compressed air.

2. Bearings

   A. Clean all bearings as directed in Subject 20-30-01, "Cleaning and Relubricating Antifriction Bearings."

INSPECTION/CHECK

1. Visual

   A. Examine all metal parts for pits, scratches, cracks, corrosion, and damage. Use a strong light and 10-power magnification.

   B. Examine entire basic assembly for corrosion, damage, and general condition of paint and finish.

   C. Examine all bearings for excessive radial or axial play.

   D. Examine all bushing and boltholes for eccentric or excessive wear.

   E. Examine all bushings for eccentric or excessive wear.

2. Special Check

   A. If visual examination discloses evidence of defects in any of listed parts, perform following checks to verify defect exists.

      (1) Dye Penetrant

         (a) Link (6, Fig. 1102)

         (b) Crank (30, Fig. 1102)

         (c) Tailgate (5, Fig. 1103)

         (d) Flap (15, Fig. 1101)

      (2) Magnetic Particle

         (a) Hinge Fitting (5, 13, or 21, Fig. 1104)

REPAIR

1. Repair

A. Remove corrosion and minor defects from metal parts by polishing lightly with abrasive cloth, 200 grit or finer. Do not exceed limits given in Fig. 601. Refinish polished area as necessary for protection against corrosion.

B. Remove minor surface defects from the flap actuator lug by blending and polishing. One edge of each lug half may be worn to limits shown in Fig. 401. Refinish polished areas, as necessary, for protection against corrosion.

2. Refinish

NOTE: Refer to 20-30-02 for stripping of protective finishes, and to 20-41-01 for explanation of F and SRF finish codes.

A. Deleted

(1) Link assembly (5, Fig. 1102) -- Apply SRF-2.30 followed by SRF-14.9814 all over except on bearing surfaces. Apply SRF-14.9815-701 on 2-inch area at end of link with hole for cylindrical nut. Material: Al Alloy.

(2) Crank (30, Fig. 1102) -- Refinish as shown in Fig. 402. Material: Al Alloy.

(3) Tailgate (6, Fig. 1103) -- Apply F-2.20 followed by SRF-12.205 all over except in holes. Material: Al Alloy.

(4) Tailgate assembly (5, Fig. 1103)

(a) P/N 69-37808-1, -2 -- Apply SRF-12.63 all over except on bushings. Material: Al Alloy.

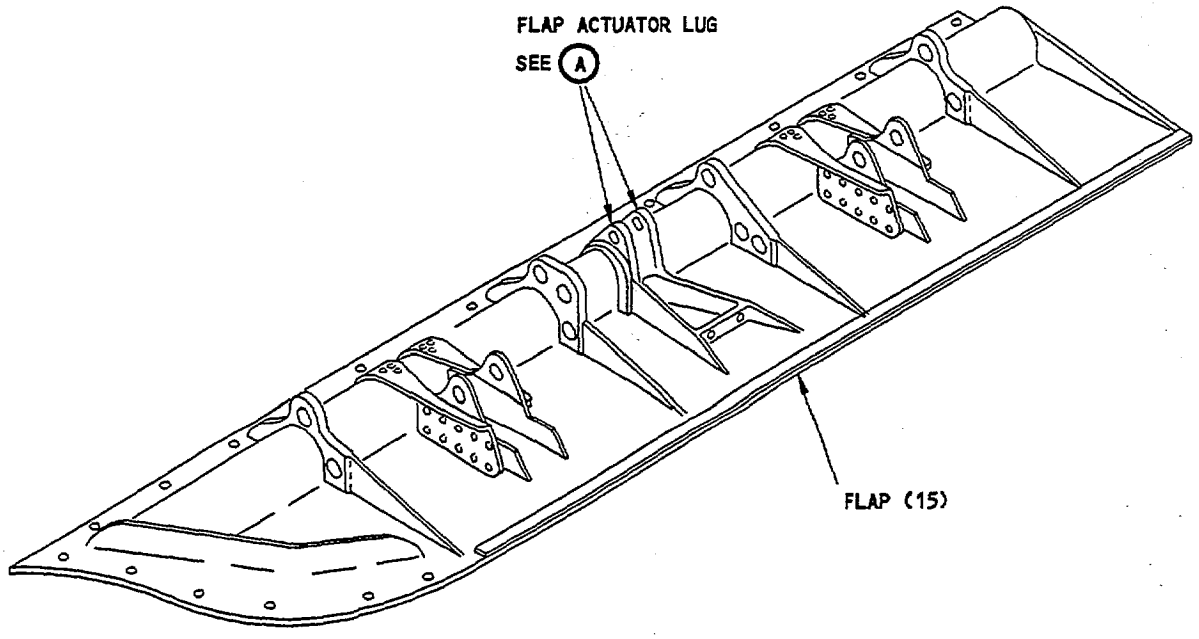
(b) P/N 69-37808-9, -10 -- Apply SRF-14.967 all over except on bushings. Material: Al Alloy.

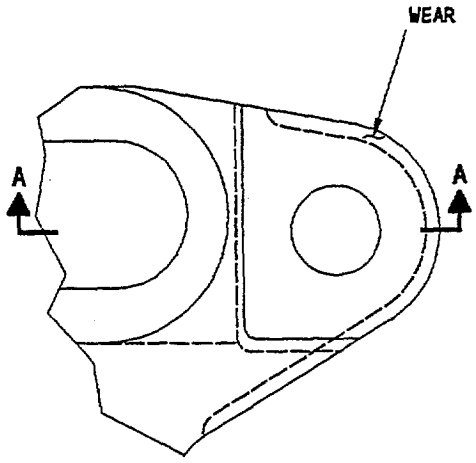
(5) Lug (16, 19, or 26, Fig. 1103) -- Apply SRF-2.30 all over except in bushing or bearing bore. Material: Al Alloy.

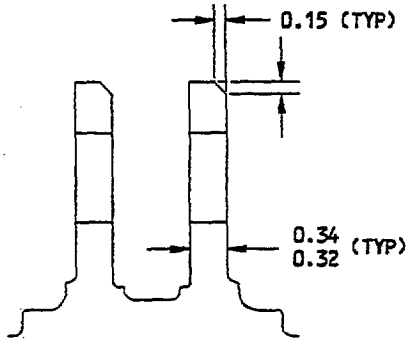
(6) Lug assembly (15, 18, or 25, Fig. 1103) -- Apply SRF-14.9814 or SRF-14.967 all over except on bearing or bushings. Material: Al Alloy.

(7) Hinge fitting (5, 13, or 21, Fig. 1104) -- Apply SRF-1.285 all over except omit primer in bore. Material: Steel, AMS 4340, 180-200 ksi.

(8) Hinge assembly (4, 12, or 20, Fig. 1104) -- Apply SRF-12.63 all over except on bearing surfaces.







SECTION A-A

Flap Actuator Lug - Allowable Wear
Figure 401

E36825

(9) Flap (15, Fig. 1101) -- Apply F-2.20 followed by SRF-12.205 all over except in holes. Material: Al Alloy.

(10) Flap assembly (14, Fig. 1101)

(a) P/N 65-49532-1, -2, -5, -6, -9, and -10 -- Apply SRF-14.9814 all over except on bushings.

(b) P/N 65-49532-17, -18 -- Apply SRF-14.967 all over except on bushings.

(c) P/N 65-49532-21, -22, -23, -24, -- Apply F-14.9864-707 to exterior surfaces. Apply SRF-14.967 (F-14.9864-707 optional) to interior surface. Mask all bushings.

(d) Use primer BMS 10-79, Type 2 (F-19.46) under decorative surfaces. BMS 10-79, Type 2 is the preferred option to primer BMS 10-11, Type 1.

B. Coat all seal retainers (3, 8, 9, 10, and 11, Fig. 1101) with primer BMS 10-11, Type 1.

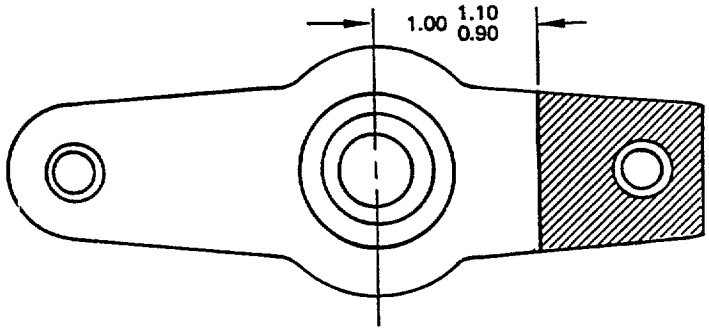
3. Replacement

A. Replace all parts worn or damaged beyond simple repair.

B. Replace all cotter pins at each overhaul.

C. Replace all rubber or fabric seals at each overhaul.

D. Replace rub strip (19, Fig. 1101) at each overhaul. Bond new rub strip in place as directed in 20-50-12, Application of Adhesives, using type 38, special method 2.



SRF-149814 ALL OVER FOLLOWED BY
SRF-14.9815-701 ALL OVER SHADED AREA
NO FINISH IN HOLES

Finishing Data
Figure 402

E. Replace defective bearing (11 or 31, Fig. 1102; 27, Fig. 1103; 6, 14, or 22, Fig. 1104) as follows:

(1) Remove old bearing. Coat new bearing with a light film of grease, MIL-G-23827. Install new bearing and roller swage in place per 20-50-03.

F. Replace defective bushings as follows:

(1) Press old bushing out of housing with a mandrel.

(2) Coat faying surfaces of bushing and housing with primer BMS 10-11, Type 1. Press bushing in to housing while primer is wet.

(3) Bushings (16, 17 and 18, Fig. 1101) — after installation apply a fillet seal around flanges, using BMS 5-95 (BMS 5-79 optional) per 20-50-11, Application of Aero Dynamic smoothing sealant, for corrosion protection.

G. Replace damaged shims (10 or 28), Fig. 1103; and 7, 15, 16, 23 or 24), Fig. 1104 as follows:

(1) Adjust thickness of basic shim to match damaged shim by removing 0.003-inch laminations.

(2) Drill through shim to match holes in adjacent parts.

(3) Deburr and coat shim with primer BMS 10-11, Type 1.

H. Replace flap (15, Fig. 1101) or tailgate (6, Fig. 1103) as follows:

NOTE: Flap and tailgate become a matched set when drilled for attaching parts during initial assembly.

(1) Replacement of either part requires drilling the new part, using the mating part and master tooling, or by drilling the new part using the old part as a pattern.

(2) It is recommended that assemblies be returned to Boeing for repair when replacement of either a flap or a tailgate is required.

4. Materials

NOTE: Use listed material or equivalent substitute.

A. Primer — BMS 10-11, Type 1

B. Grease — MIL-G-23827

C. Sealant — BMS 5-95

D. Primer -- BMS 10-79, Type 2

ASSEMBLY

1. Buildup

A. Installation of Aerodynamic Seals (Fig. 1101)

NOTE: Exact positioning of seals is done on the aircraft.

(1) Tailgate Seals (Fig. 1103)

(a) Attach seal (32) with fasteners (29, 30, and 31).

(2) Flap Seals (Fig. 1101)

(a) Attach seals (12 and 13) and retainers (8, 9, 10, and 11) with bolts (7) and nuts (6).

(b) Attach seals (4 and 5) and retainer (3) with bolts (2) and nuts (1).

B. Installation of Hinges (Fig. 1104)

(1) Locate hinge assemblies (12 and 20) and fit shims (15, 16, 23 and 24) in location from which disassembled. Adjust shims by removing 0.003-inch laminations as necessary to eliminate gap between hinge and structure, and to maintain 9.305- to 9.325-inch dimension as shown in Fig. 501. Drill shims to match holes in adjacent parts, deburr and coat with primer BMS 10-11, Type 1.

NOTE: Maximum allowable gap is 0.005 inch. Maximum shim thickness is 0.21 inch; total shim thickness, each hinge, is 0.21 inch.

(2) Coat bolts (11 and 19) with primer BMS 10-11, Type 1. Install hinge assemblies while primer is wet, with bolts (11 and 19), washers (10 and 18) and nuts (9 and 17).

(3) Locate inboard hinge assembly (4), and fit shims (7) in location from which disassembled to eliminate gap between hinge and flap structure. Adjust thickness of shims by removing 0.003-inch laminations. Drill holes through shims to match holes in adjacent parts, deburr and coat shims with primer BMS 10-11, Type 1.

NOTE: Maximum allowable gap is 0.005 inch. Maximum shim thickness one side is 0.09 inch; maximum total shim thickness is 0.09 inch.

(4) Locate sensor actuator (8). Coat two bolts (3) with primer BMS 10-11, type 1, and attach actuator and inboard hinge assembly using washers (2) and nuts (1), while primer is wet.

(5) Locate outboard hinge assembly (4), and fit shims (7) in location from which disassembled to eliminate gap between hinge and flap structure. Adjust thickness of shims by removing 0.003-inch laminations. Drill holes through shims to match adjacent parts. deburr and coat shims with primer, BMS 10-11, type 1.

NOTE: Maximum allowable gap is 0.005 inch, maximum shim thickness one side is 0.09 inch; maximum total shim thickness is 0.09 inch.

(6) Coat two bolts (3) with primer BMS 10-11, type 1. Install hinge assembly (4) with bolts (3), washers (2) and nuts (1). while primer is wet.

CAUTION: IF ANY HINGE FITTING HAS BEEN REPLACED, EXISTING HINGE FITTINGS SHOULD BE USED AS A REFERENCE TO MAINTAIN INTEGRITY OF FLAP HINGE CENTERLINE.

C. Installation of Tailgate and Lug Assemblies (See figure 1103.)

(1) Locate lug assemblies (25) in the positions noted during disassembly and adjust by shims (28) to centerline dimensions shown in figure 501. Coat bolts (23 and 24) with primer, BMS 10-11, type 1, and install lug assemblies while primer is wet, using washers (22) and nuts (21).

(2) Coat bolts (14) with corrosion preventive compound, MIL-C-11796, class 3, and attach lug assemblies (15 and 18) to lug assemblies (25) using washers (13), nuts (12), and cotter pins (11).

(3) Locate shims (10) between lug assemblies (15 and 18) and tailgate assembly (5). Adjust thickness of shims, by removing 0.003-inch laminations, to position tailgate to dimensions shown in figure 501. Coat shims with primer, BMS 10-11, type 1, after delamination. Coat bolts (3 and 4) with primer, BMS 10-11, type 1, and attach tailgate assembly to lug assemblies using washers (2) and nuts (1).

D. Installation of Linkage (See figure 1102.)

(1) Locate crank assemblies (29). Install bolts (28), washers (27), nuts (26) with corrosion preventive compound, MIL-C-11796, class 3.

(2) Tighten nuts (26) within torque range of 5 to 10 pound-inches, and install cotter pins (25).

(3) Assemble rods (22 and 22A), checknuts (23 and 23A), and rod end bearings (24 and 24A) to make up two rod assemblies (21).

(4) Adjust length of each rod assembly (21) to 5.24 inches between centerlines of rod end bearings. Tighten checknuts against rod end bearings.

CAUTION: RODS (22A), CHECKNUTS (23A) AND ROD END BEARINGS (24A) HAVE LEFT-HAND THREADS.

NOTE: Do not wirelock rod assemblies (21). Final adjustment and wirelocking to be accomplished on installation of flap assembly on airplane.

(5) Coat bolts (15 and 19) with corrosion preventive compound MIL-C-11796 class 3. Install rod assemblies (21) with clamp-up bushings (20), bolts (15 and 19), washers (14 and 18), nuts (13 and 17), and cotter pins (12 and 16).

(6) Install cylindrical nut (10), washer (9), checknut (7), and bearing (8) on each basic link assembly (5).

NOTE: Install spacers (33) between link and crank on inboard link assembly only.

(7) Adjust length of each link assembly (5) to 8.30 inches between centerlines of end bearings. Tighten checknuts against link.

NOTE: Do not wirelock link assemblies (5). Final adjustment and wirelocking to be accomplished on installation of flap assembly on airplane.

(8) Coat bolts (4) with corrosion preventive compound, MIL-C-11796, class 3, and install link assemblies (5) in flap using washers (3), nuts (2), and cotter pins (1). Match black painted end of link with black painted end of crank (30).

E. Check that tailgate operates freely and contacts flap pads and rub strip as shown in Fig. 501. If necessary, grind off high spots on surface of rub strip to make firm contact.

F. Touch up finish as necessary on entire assembly after completion of buildup.

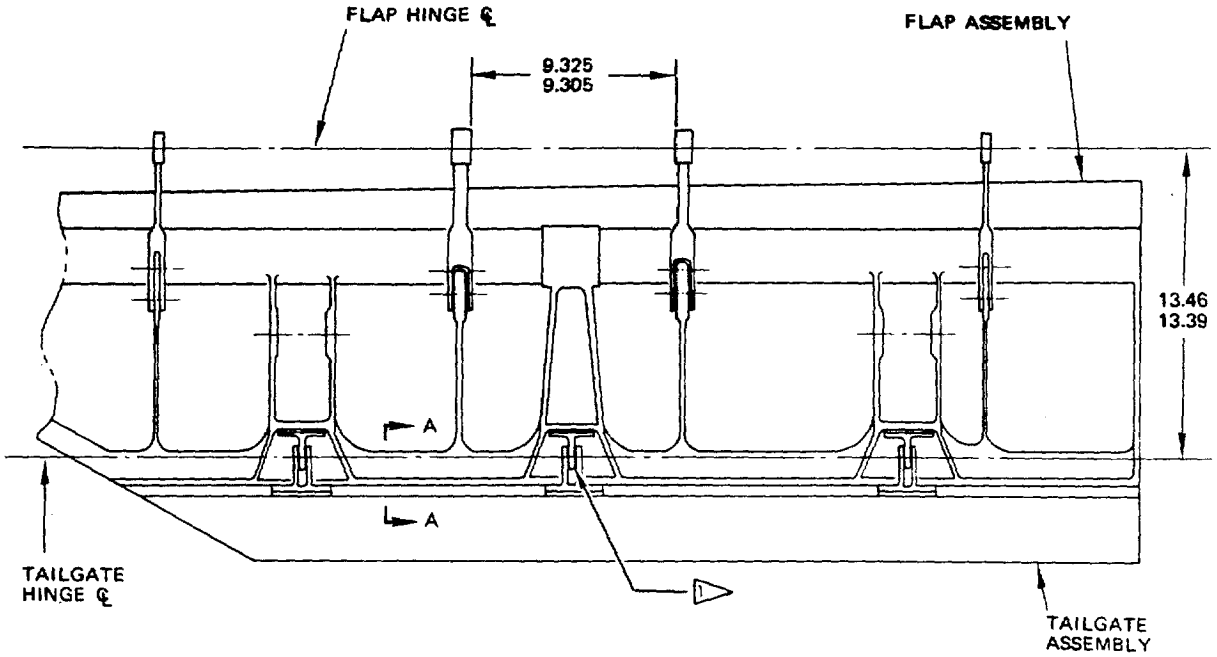
G. For convenience of installation, install bolt (54), pin (57), washer (55) and nut (56) in link assembly (5). Install spacers (58) on inboard link assembly only.

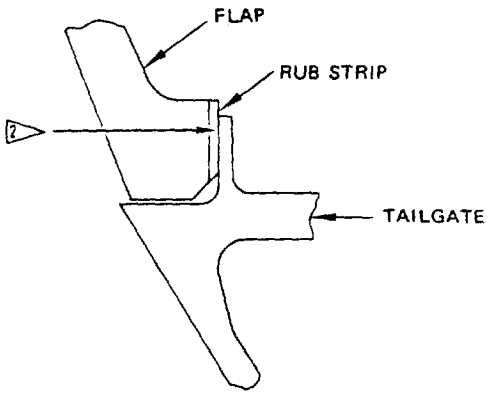
2. Materials

NOTE: Use listed materials or equivalent substitutes.

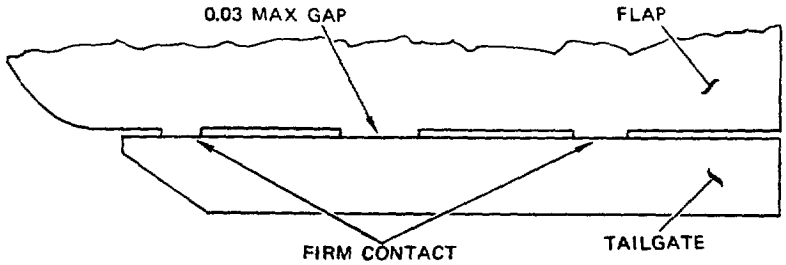
A. Primer -- BMS 10-11, type 1

B. Corrosion Preventive Compound -- MIL-C-11796, Class 3





SECTION A–A



1 THIS HINGE TO BE IN LINE WITH END HINGES WITHIN 0.02 T.I.R.

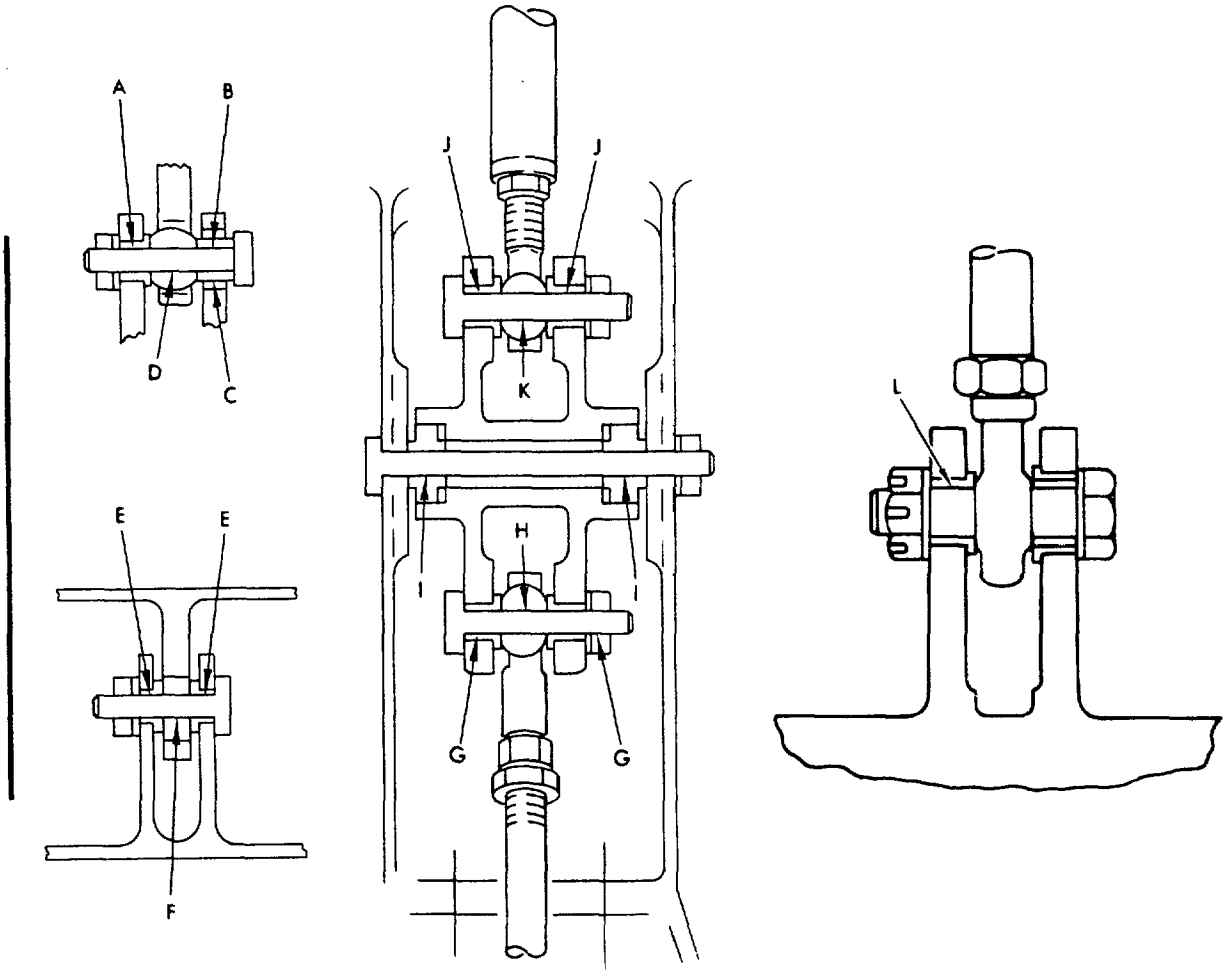
2 TAILGATE ASSEMBLY SHALL FIRMLY CONTACT RUB STRIP A MINIMUM OF 25% OF SURFACE AREA

NOTE: ALL DIMENTIONS ARE IN INCHES

FITS AND CLEARANCES

1. The fits and clearances table lists design dimensions and service wear limits for close tolerance parts of the assembly that are subject to wear or corrosion. Unless otherwise specified, parts should be returned to the design dimensions whenever rework is accomplished.

2. Clearances are given to aid assembly of the components. The values given in the Maximum Allowable Clearance column are the maximum permitted to ensure proper functioning of the unit. If assembled parts fail to meet this requirement, one or more of the parts must be rejected. Parts that are rejected should be reworked if within the rework limits given in the Repair procedure; if not within rework limits, the parts should be scrapped. It is recommended that the design clearances be used as the guiding assembly criteria when newly reworked parts are assembled.



Fits and Clearances
Figure 601 (Sheet 1)

Ref Letter Fig.601	Mating Item No. Fig.	Design Dimensions: Dimensions (inches) Min	Dimensions (inches) Max	Assembly Clearance (inch) Min	Assembly Clearance (inch) Max	Service Wear Limits: Dimension Limits (inches) Min	Dimension Limits (inches) Max	Maximum Allowable Clearance (inch)
A	ID *[1]	0.2500	0.2515	0.0005	0.0025	0.2450	0.2545	0.0050
	OD 1102-19	0.2490	0.2495					
B	ID 1102-20	0.2495	0.2510	0.0000	0.0020	0.2455	0.2535	0.0040
	OD 1102-19	0.2490	0.2495					
C	ID 1103-9	0.4375	0.4390	0.0020	0.0055	0.4285	0.4445	0.0090
	OD 1102-20	0.4335	0.4355					
D	ID *[2]	0.2495	0.2500	0.0000	0.0010	0.2475	0.2515	0.0020
	OD 1102-19	0.2490	0.2495					
E	ID *[3]	0.2500	0.2515	0.0005	0.0025	0.2450	0.2545	0.0050
	OD 1103-14	0.2490	0.2495					
F	ID 1103-27	0.2495	0.2500	0.0000	0.0010	0.2475	0.2515	0.0020
	OD 1103-14	0.2490	0.2495					
G	ID 1102-32	0.2500	0.2515	0.0005	0.0025	0.2450	0.2545	0.0050
	OD 1102-15	0.2490	0.2495					
H	ID *[2]	0.2495	0.2500	0.0000	0.0010	0.2475	0.2515	0.0020
	OD 1102-15	0.2490	0.2495					
I	ID 1102-31	0.2495	0.2500	0.0000	0.0010	0.2475	0.2515	0.0020
	OD 1102-28	0.2490	0.2495					

Fits and Clearances
Figure 601 (Sheet 2)

Ref Letter Fig.601	Mating Item No. Fig.	Design Dimensions				Service Wear Limits		
		Dimensions (inches)		Assembly Clearance (inch)		Dimension Limits (inches)		Maximum Allowable Clearance (inch)
		Min	Max	Min	Max	Min	Max	
J	ID 1102-32	0.2500	0.2515	0.0005	0.0025		0.2545	0.0050
	OD 1102-4	0.2490	0.2495			0.2450		
K	ID 1102-8	0.2495	0.2500	0.0000	0.0010		0.2515	0.0020
	OD 1102-4	0.2490	0.2495			0.2475		
L	ID 1101-16	0.6250	0.6265	0.0010	0.0035		0.6310	0.0070
	OD Ref	0.6230	0.6240			0.6180		

*[1] Items 1103-7 and -8

*[2] Items 1102-24 or -24A

*[3] Items 1103-17 and -20

STORAGE INSTRUCTIONS

1. Coat all bushing and bearing surfaces with grease, Specification MIL-G-23827.

2. Wrap entire assembly in nonabsorbent material, and store in a cool, dry area, preferably humidity controlled. Place package where it will not be moved frequently, or handled roughly.

3. For further storage instructions, refer to Subject 20-44-02, "Temporary Protective Coatings," and Subject 20-70-01, "Protection, Storage, and Handling of Airplane Components."

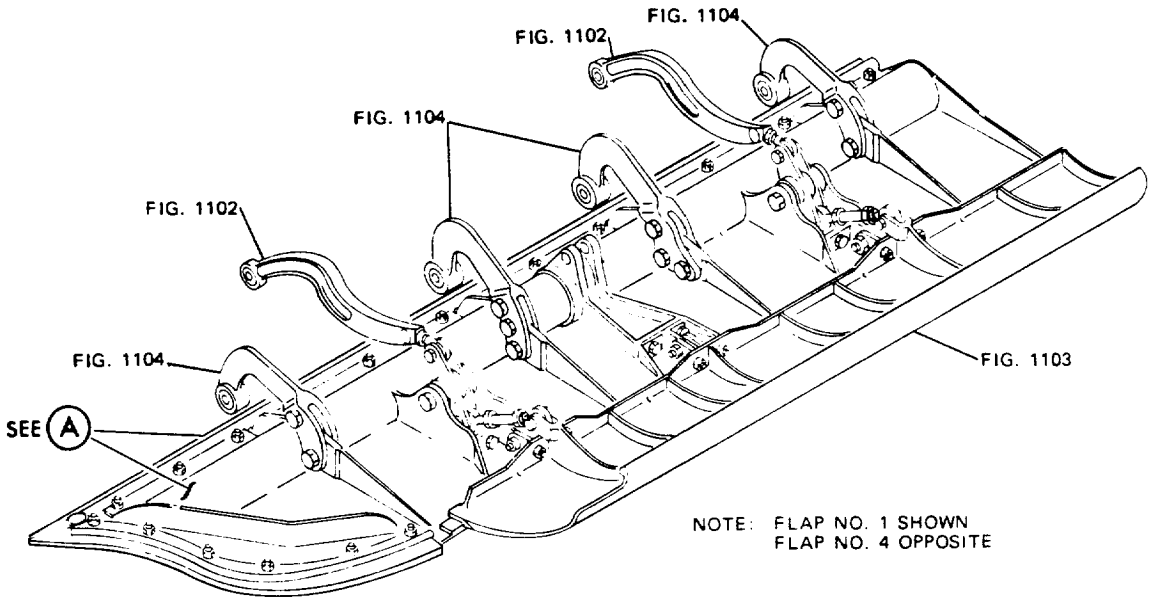
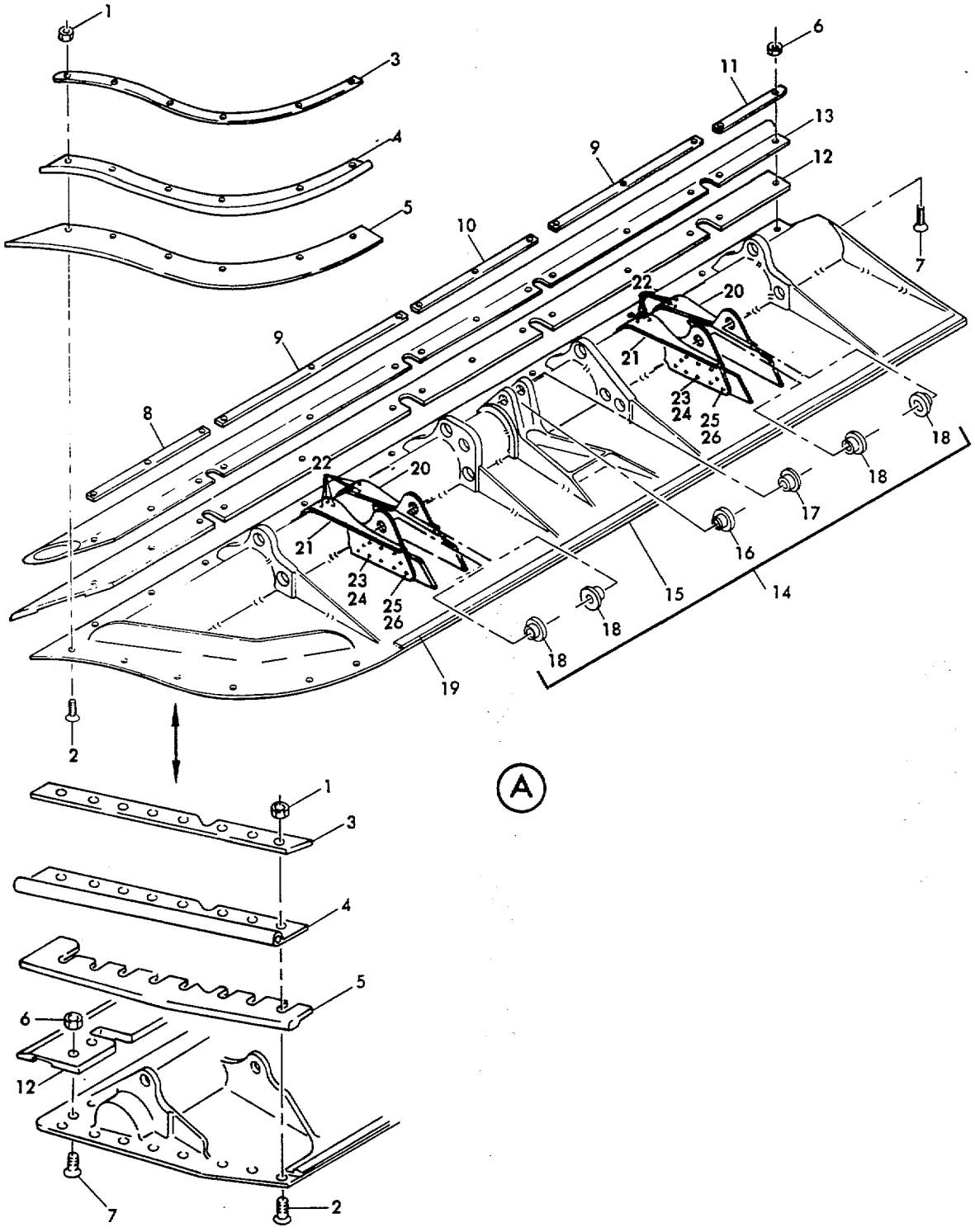


FIG. 1104
FIG. 1102
FIG. 1104
FIG. 1102
FIG. 1104
SEE (A)
FIG. 1103
NOTE: FLAP NO. 1 SHOWN
FLAP NO. 4 OPPOSITE



Leading Edge Flap Assemblies No. 1 and No. 4
Figure 1101 (Sheet 2)

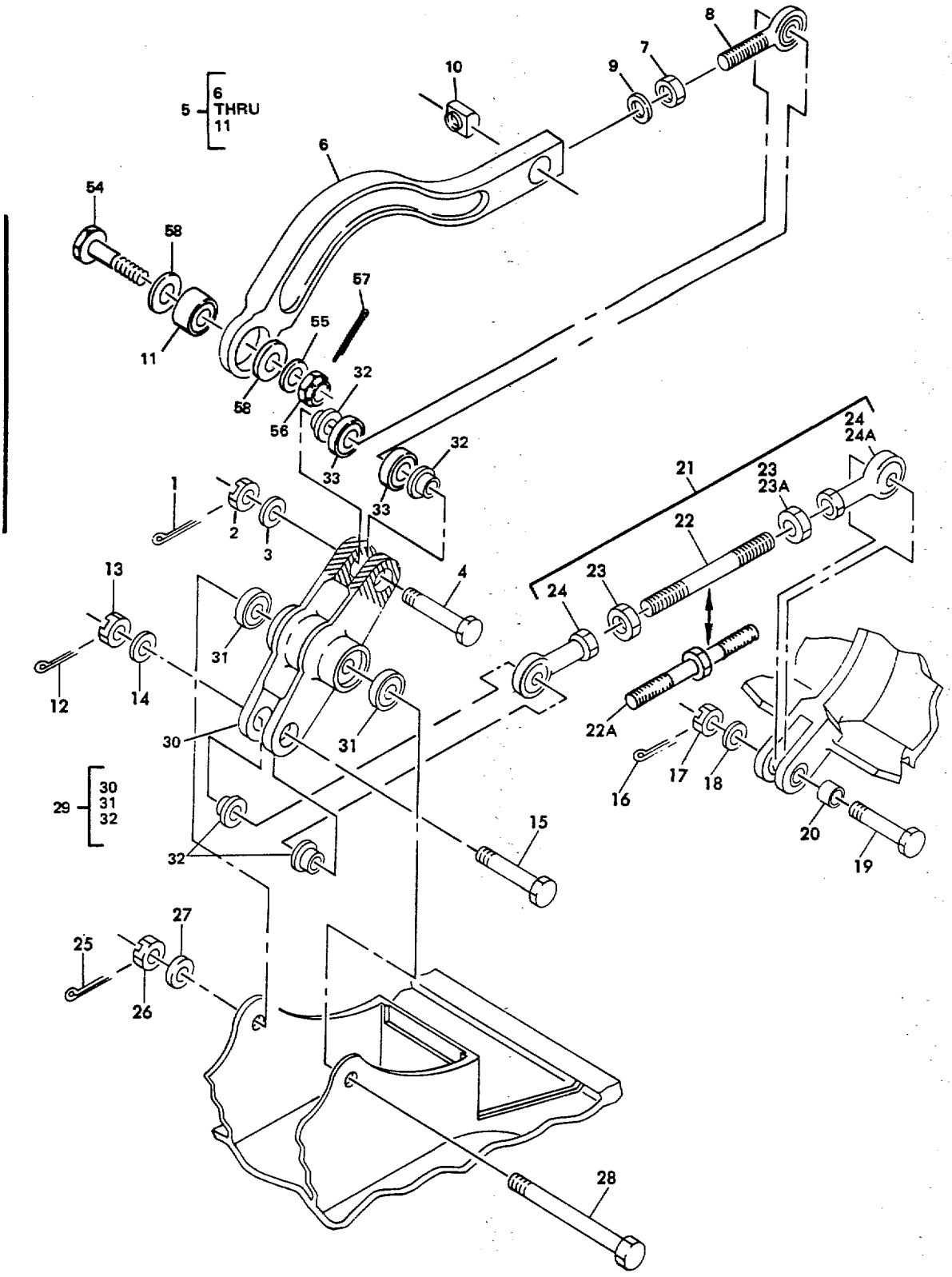
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1101-					
	65-46424-1		LEADING EDGE FLAP ASSY NO. 1	A	RF
	65-46424-5		LEADING EDGE FLAP ASSY NO. 1	B	RF
	65-46424-7		LEADING EDGE FLAP ASSY NO. 1	C	RF
	65-46424-9		LEADING EDGE FLAP ASSY NO. 1	D	RF
	65-46424-13		LEADING EDGE FLAP ASSY NO. 1	E	RF
	65-46424-19		LEADING EDGE FLAP ASSY NO. 1	F	RF
	65-46424-21		LEADING EDGE FLAP ASSY NO. 1 (SB 57-1077)	G	RF
	65-46424-23		LEADING EDGE FLAP ASSY NO. 1	H	RF
	65-46424-25		LEADING EDGE FLAP ASSY NO. 1	I	RF
	65-46424-27		LEADING EDGE FLAP ASSY NO. 1	J	RF
	65-46424-29		LEADING EDGE FLAP ASSY NO. 1	K	RF
	65-46424-2		LEADING EDGE FLAP ASSY NO. 4	N	RF
	65-46424-6		LEADING EDGE FLAP ASSY NO. 4	O	RF
	65-46424-8		LEADING EDGE FLAP ASSY NO. 4	P	RF
	65-46424-10		LEADING EDGE FLAP ASSY NO. 4	Q	RF
	65-46424-14		LEADING EDGE FLAP ASSY NO. 4	R	RF
	65-46424-20		LEADING EDGE FLAP ASSY NO. 4	S	RF
	65-46424-22		LEADING EDGE FLAP ASSY NO. 4 (SB 57-1077)	T	RF
	65-46424-24		LEADING EDGE FLAP ASSY NO. 4	U	RF
	65-46424-26		LEADING EDGE FLAP ASSY NO. 4	V	RF
	65-46424-28		LEADING EDGE FLAP ASSY NO. 4	W	RF
	65-46424-30		LEADING EDGE FLAP ASSY NO. 4	X	RF

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1101					
	65-67101-3		. SEAL INST	ABC	1
	65-67101-4		. SEAL INST (OPP)	NDP	1
	65-67101-23		. SEAL INST	E	1
	65-67101-24		. SEAL INST (OPP)	R	1
	65-67101-29		. SEAL INST	DF-K	1
	65-67101-30		. SEAL INST (OPP)	QS-X	1
1	BACN10JC3		. . NUT (USED ON 65-67101-3,-4,-23, -24)		6
1	BACN10JC3		. . NUT (USED ON 65-67101-29, -30)		8
2	BACB30LU3-4		. . BOLT (USED ON 65-67101-3, -4, -23, -24)		6
2	BACB30LU3-4		. . BOLT (USED ON 65-67101-29, -30)		8
3	69-54940-1		. . RETAINER (USED ON 65-67101-3, -4, -23, -24)		1
3	69-62784-1		. . RETAINER (USED ON 65-67101-29, -30)		1
4	65-67101-5		. . SEAL (USED ON 65-67101-3, -23)		1
4	65-67101-6		. . SEAL (USED ON 65-67101-4, -24)		1
4	69-62783-1		. . SEAL (USED ON 65-67101-29)		1
4	69-62783-2		. . SEAL (USED ON 65-67101-30)		1
5	69-54935-3		. . SEAL *[4] (USED ON 65-67101-3)		1
5	69-54935-4		. . SEAL *[4] (USED ON 65-67101-4)		1
5	69-54935-5		. . SEAL (SB 57-1028)(USED ON 65-67101-3, -23)		1
5	69-54935-6		. . SEAL (SB 57-1028)(USED ON 65-67101-4, -24)		1
5	69-62782-1		. . SEAL (USED ON 65-67101-29)		1
5	69-62783-2		. . SEAL (USED ON 65-67101-30)		1
6	BACN10JC3		. . NUT (USED ON 65-67101-3, -4, -23, 24)		14
6	BACN10JC3		. . NUT (USED ON 65-67101-29, -30)		13
7	BACB30LU3-6		. . BOLT (USED ON 65-67101-3, -4)		14
7	BACB30LU3-5		. . BOLT (USED ON 65-67101-23, -24)		14
7	BACB30LU3-5		. . BOLT (USED ON 65-67101-29, -30)		13
8	69-43513-23		. . RETAINER (REPLS 69-43513-14) (USED ON 65-67101-3, -4)		1
9	69-43513-24		. . RETAINER (REPLS 69-43513-15 (USED ON 65-67101-3, -4)		2
10	69-43513-25		. . RETAINER (REPLS 69-43513-16) (USED ON 65-67101-3, -4)		1
11	69-43513-26		. . RETAINER (REPLS 69-43513-17) (USED ON 65-67101-3, -4)		1
12	69-54943-7		. . SEAL *[4] (USED ON 65-67101-3)		1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1101					
12	69-54943-8		. . SEAL *[4] (USED ON 65-67101-4)		1
12	69-54943-3		. . SEAL *[4] (USED ON 65-67101-3)		1
12	69-54943-4		. . SEAL *[4] (USED ON 65-67101-4)		1
12	69-54943-11		. . SEAL *[4] (USED ON 65-67101-3, -23)		1
12	69-54943-12		. . SEAL *[4] (USED ON 65-67101-4, -24)		1
12	69-54943-29		. . SEAL (PREF)(USED ON 65-67101-29)		1
12	69-54943-23		. . SEAL (OPT TO 69-54943-29)		1
12	69-54943-30		. . SEAL (PREF)(USED ON 65-67101-30)		1
12	69-54943-24		. . SEAL (OPT TO 69-54943-30)		1
13	69-43512-13		. . SEAL (USED ON 65-67101-3)		1
13	69-43512-14		. . SEAL (USED ON 65-67101-4)		1
14	65-49532-5		. FLAP ASSY (REPLS 69-54942-1 ON 65-46424-1)*[1]	AB	1
14	69-54942-1		. FLAP ASSY (REPLD BY 65-49532-5)	A	1
14	65-49532-6		. FLAP ASSY (REPLS 69-54942-2 ON 65-46424-2)*[1]	NO	1
14	69-54952-2		. FLAP ASSY (REPLD BY 65-49532-6)	N	1
14	65-49532-9		. FLAP ASSY	CE	1
14	65-49532-10		. FLAP ASSY	PR	1
14	65-49532-17		. FLAP ASSY	D	1
14	65-49532-18		. FLAP ASSY	Q	1
14	65-49532-21		. FLAP ASSY	FG	1
14	65-49532-22		. FLAP ASSY	ST	1
14	65-49532-23		. FLAP ASSY	HJ	1
14	65-49532-24		. FLAP ASSY	UW	1
14	65-49532-27		. FLAP ASSY	I	1
14	65-49532-28		. FLAP ASSY	V	1
14	65-49532-31		. FLAP ASSY	K	1
14	65-49532-32		. FLAP ASSY	X	1
15	65-49532-7		. . FLAP (USED ON 65-49532-5)		1
15	65-49532-8		. . FLAP (USED ON 65-49532-6)		1
15	65-49532-11		. . FLAP (USED ON 65-49532-9)		1
15	65-49532-12		. . FLAP (USED ON 65-49532-10)		1
15	65-49532-19		. . FLAP (USED ON 65-49532-17,-21)		1
15	65-49532-20		. . FLAP (USED ON 65-49532-18,-22)		1
15	65-49532-3		. . FLAP (USED ON 69-54942-1)		1
15	65-49532-4		. . FLAP (USED ON 69-54942-2)		1
15	65-49532-25		. . FLAP (USED ON 65-49532-23)		1
15	65-49532-26		. . FLAP (USED ON 65-44532-24)		1
15	65-49532-29		. . FLAP (USED ON 65-49532-27)		1
15	65-49532-30		. . FLAP (USED ON 65-49532-28)		1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1101					
15	65-49532-33		. . FLAP (USED ON 65-49532-31)		1
15	65-49532-34		. . FLAP (USED ON 65-49532-32)		1
16	BACB28X10C33		. . BUSHING		1
17	BACB28X14B33		. . BUSHING		1
18	BACB28X4B19		. . BUSHING		4
19	65-46424-3		. STRIP, RUB *[2]	ABCE NDPR	1
19	65-46424-15		. STRIP, RUB *[2]	DF-K QS-X	1
20	65-86048-2		. TEE (SB 57-1077) *[3]	A-G	2
20	65-86048-1		. TEE (SB 57-1077) *[3]	N-T	2
21	65-86048-1		. TEE (SB 57-1077) *[3]	A-G	2
21	65-86048-2		. TEE (SB 57-1077) *[3]	N-T	2
22	BACB30LB6-5		. BOLT (SB 57-1077) *[3]	A-G N-T	20
23	BACB30FM6-3		. BOLT (SB 57-1077) *[3]	A-G N-T	32
24	BACC30M6		. COLLAR (SB 57-1077) *[3]	A-G N-T	32
25	BACB30FM5-3		. BOLT (SB 57-1077) *[3]	A-G N-T	8
26	BACC30M5		. COLLAR *[3]	A-G N-T	8

*[1] FLAP ASSEMBLIES 69-54942-1 AND -2 ARE 65-49532-1 AND -2 REWORKED AND REIDENTIFIED BY DRAWING 69-54942
*[2] RUB STRIP (19) MAY BE MADE IN TWO PIECES (MINIMUM LENGTH TO BE 10.0 INCHES)
*[3] PART OF BASIC STRUCTURE
*[4] LIMITED USE

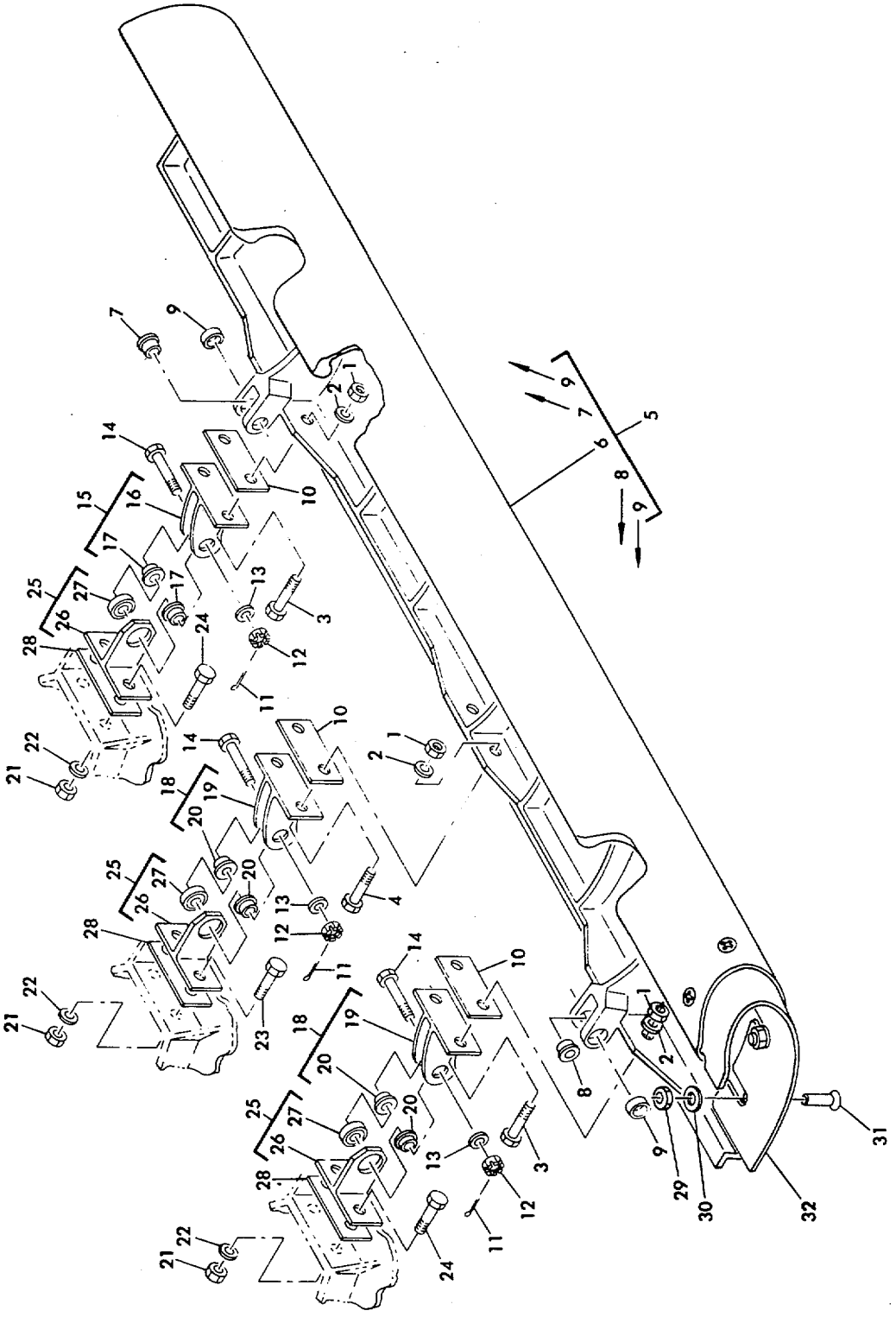


Linkage
Figure 1102

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1102-			USE CODES ASSIGNED IN FIG. 1101 LINKAGE		
1	MS24665-134		. PIN, COTTER		2
2	BACN10JD104		. NUT (REPLS AN320-4)		2
3	AN960PD416L		. WASHER		2
4	BACB30LJ4DU18		. BOLT		2
5	69-37841-11		. LINK ASSY (PREF)		2
5	69-37841-7		. LINK ASSY (SPSDS 69-37841-1) (OPT TO 69-37841-11)	A-E N-R	2
6	69-37841-13		. . LINK (USED ON 69-37841-11)		1
6	69-37841-9		. . LINK (USED ON 69-37841-7)		1
7	NAS509-5		. . CHECKNUT		1
8	REM8ATC10-6		. . BEARING, V21335 (BOEING 10-60779-124)		1
8	DREM4-292		. . BEARING, V81376 (BOEING 10-60779-124)		1
8	YTM-187		. . BEARING, V77896 (BOEING 10-60779-124)		1
8	177144		. . BEARING, V09455 (BOEING 10-60779-124)		1
9	AN960PD516L		. . WASHER		1
10	BACN10CP5L		. . NUT, CYLINDRICAL		1
11	MS21232-4		. . BEARING		1
12	MS24665-134		. PIN, COTTER		2
13	BACN10JD104		. NUT (REPLS AN320-4)		2
14	AN960PD416L		. WASHER		2
15	BACB30LJ4DU18		. BOLT		2
16	MS24665-134		. PIN, COTTER		2
17	BACN10JD104		. NUT (REPLS AN320-4)		2
18	AN960PD416L		. WASHER		2
19	BACB30LJ4DU16		. BOLT		2
20	NAS74A4-005P		. BUSHING, CLAMPUP		2
21	69-61262-3		. ROD ASSY (PREF)		2
21	69-37244-3		. ROD ASSY (OPT)		2
22	NAS354-5-310		. . ROD *[2]		1
22A	69-61262-8		. . ROD *[1]		1
23	NAS509-5		. . CHECKNUT *[2]		2
23	NAS509-5		. . CHECKNUT *[1]		1
23A	NAS509L5		. . CHECKNUT *[1]		1

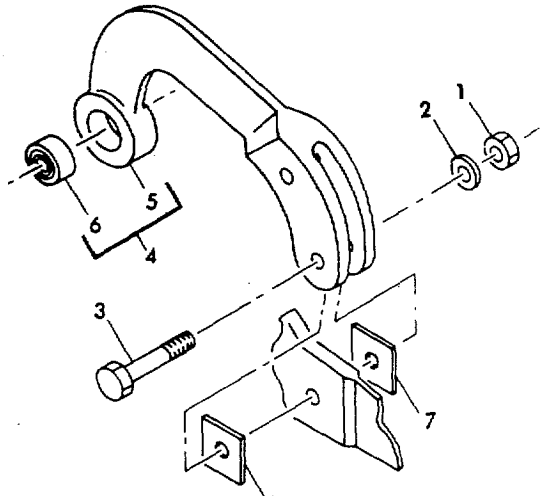
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1102-24	REF8ATC10		. . BEARING, ROD END, V21335 (BOEING 10-60779-201)		*[3]
24	YTF124A		. . BEARING, ROD END , V77896 (BOEING 10-60779-201)		*[3]
24	DREF4-055		. . BEARING, ROD END, V81376 (BOEING 10-60779-201)		*[3]
24	02-727-0250		. . BEARING, ROD END, V09455 (BOEING 10-60779-201)		*[3]
24A	REFL8ATC10		. . BEARING, ROD END, V21335 (BOEING 10-60779-201L)*[1]		1
24A	YTF124AL		. . BEARING, ROD END, V77896 (BOEING 10-60779-201L)*[1]		1
24A	DREFLH4-055		. . BEARING, ROD END, V81376 (BOEING 10-60779-201L)*[1]		1
24A	12-727-0250		. . BEARING, ROD END, V09455 (BOEING 10-60779-201L)*[1]		1
25	MS24665-134		. PIN, COTTER		2
26	BACN10JD104		. NUT (REPLS AN320-4)		2
27	AN960PD416L		. WASHER		2
28	BACB30LJ4DU44		. BOLT		2
29	65-49528-11		. CRANK ASSY	F-K S-X	2
29	65-49528-7		. CRANK ASSY (OPT)	A-G N-T	2
30	65-49528-8		. . CRANK (USED ON 65-49528-7)		1
30	65-49528-12		. . CRANK (USED ON 65-49528-11)		1
31	BACB10A661		. . BEARING		2
32	BACB28X4B25		. . BUSHING		4
33	NAS1515M10		. . WASHER *[4]		2
			INSTALLATION PARTS		
54	BACB30LJ4D419		BOLT		1
55	AN960PD416L		WASHER		1
56	AN320-4		NUT		1
57	MS24665-134		PIN, COTTER		1
58	69-55450-1		SPACER *[4]		2

*[1] USED ON 69-61262-3
*[2] USED ON 65-37244-3
*[3] 1 USED ON 69-61262-3, 2 USED ON 69-37244-3
*[4] INSTALL ON INBOARD LINK ASSY ONLY

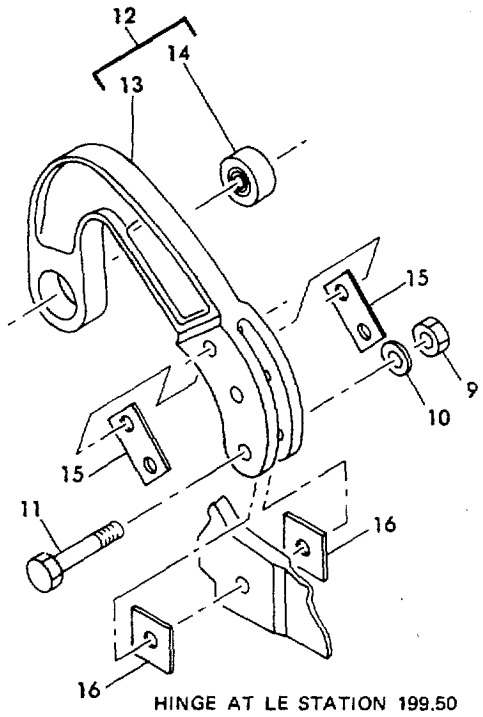


Tailgate and Lug Assemblies
Figure 1103

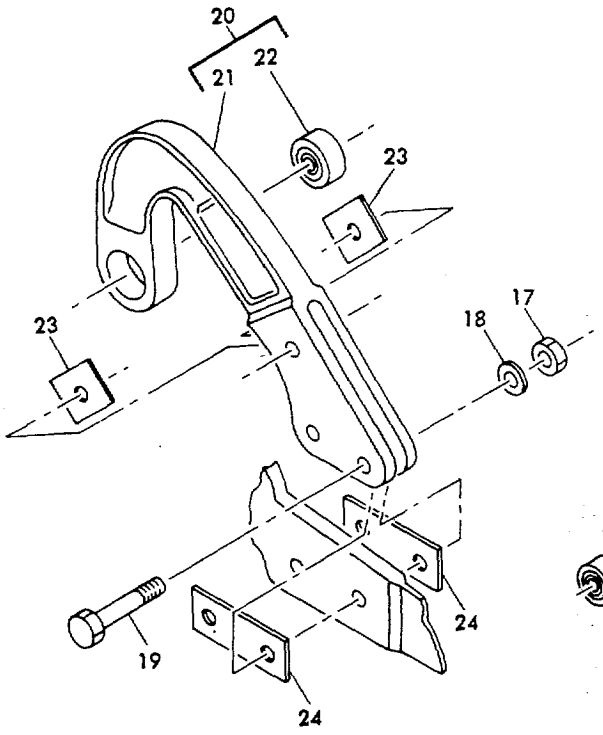
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1103-			USE CODES ASSIGNED IN FIG. 1101 TAILGATE AND LUG ASSY		
1	BACN10JC4		. NUT (REPLS NAS679A4W)		6
2	AN960PD416L		. WASHER		6
3	NAS1104-7		. BOLT		4
4	NAS1104-6		. BOLT		2
5	69-37808-1		. TAILGATE ASSY	ABCE	1
5	69-37808-2		. TAILGATE ASSY	NOPR	1
5	69-37808-9		. TAILGATE ASSY	DFGH	1
5	69-37808-10		. TAILGATE ASSY	QSTU	1
5	69-37808-13		. TAILGATE ASSY	IJK	1
5	69-37808-14		. TAILGATE ASSY	VWX	1
6	69-37808-3		. . TAILGATE (USED ON 69-37808-1)		1
6	69-37808-4		. . TAILGATE (USED ON 69-37808-2)		1
6	69-37808-11		. . TAILGATE (USED ON 69-37808-9)		1
6	69-37808-12		. . TAILGATE (USED ON 69-37808-10)		1
6	69-37808-15		. . TAILGATE (USED ON 69-37808-13)		1
6	69-37808-16		. . TAILGATE (USED ON 69-37808-14)		1
7	BACB28X4B13		. . BUSHING		1
8	BACB28X4B16		. . BUSHING		1
9	BACB28Y7B16		. . BUSHING		2
10	BACS40R10E26F		. SHIM		AR
11	MS24665-134		. PIN, COTTER		3
12	BACN10JD104		. NUT (REPLS AN320-4)		3
13	AN960PD416L		. WASHER		3
14	BACB30LJ4DU11		. BOLT		3
15	69-37842-1		. LUG ASSY		1
16	69-37842-7		. . LUG		1
17	BACB28X4B11		. . BUSHING (REPLS NAS77A4-11P)		2
18	69-37842-2		. LUG ASSY		2
19	69-37842-8		. . LUG		1
20	BACB28X4B11		. . BUSHING (REPLS NAS77A4-11P)		2
21	BACN10JC4		. NUT (REPLS NAS679A4W)		6
22	AN960PD416L		. WASHER		6
23	NAS1104-7		. BOLT		2
24	NAS1104-9		. BOLT		4
25	69-37842-6		. LUG ASSY	A-HJK N-UWX	3
25	69-37842-13		. LUG ASSY	IV	3
26	69-37842-12		. . LUG (USED ON 69-37842-6)		1
26	69-37842-14		. . LUG (USED ON 69-37842-13)		1
27	MS21232-4		. . BEARING		1
28	BACS40R10E23F		. SHIM		AR
28A	65-67101-29		. SEAL INST	DF-K	1
28A	65-67101-30		. SEAL INST	QS-X	1
29	BACN10JC3		. . NUT		5
30	BACW10P42S		. . WASHER		5
31	BACB30LU3-4		. . BOLT		5
32	69-62796-1		. . SEAL		1



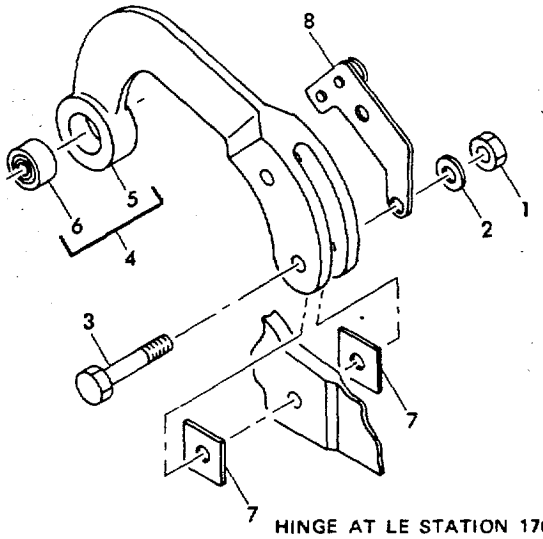
HINGE AT LE STATION 213.00



HINGE AT LE STATION 199.50



HINGE AT LE STATION 189.50



HINGE AT LE STATION 176.00

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1104-			USE CODES ASSIGNED IN FIG. 1101		
			HINGES		
1	BACN10JC4		. NUT		4
1	BACN10JC5		. NUT (USED WITH 69-37843-3 ON 65-46424-29,-30)		4
1	BACN10JC4		. NUT (USED WITH 69-37843-1,-3,-5 ON 65-46424-20 THRU -30)		4
2	AN960PD416L		. WASHER		4
2	AN960PD516L		. WASHER (USED WITH 69-37843-3 ON 65-46424-29,-30)		4
2	AN960PD416L		. WASHER (USED WITH 69-37843-1,-3,-5 ON 65-46424-20 THRU -30)		4
3	NAS1104-8		. BOLT		4
3	BACB30NF5-8		. BOLT (USED WITH 69-37843-3 ON 65-46424-29,-30)		4
3	BACB30NF4-16		. BOLT (USED WITH 69-37843-1,-3,-5 ON 65-46424-20 THRU -30)		4
4	69-37843-5		. HINGE ASSY	IV	2
4	69-37843-3		. HINGE ASSY (SB 57-1051, 1114)	B-HJK O-UWX	2
4	69-37843-1		. HINGE ASSY (PRE SB 57-1215)	AN	2
4	69-37843-3		. HINGE ASSY (POST SB 57-1215)	AN	2
5	69-37843-6		. . HINGE FTG (USED ON 69-37843-5)		1
5	69-37843-4		. . HINGE FTG (USED ON 69-37843-3)		1
5	69-37843-2		. . HINGE FTG (USED ON 69-37843-1)		1
6	MS21232-6		. . BEARING		1
7	BACS40R10D10F		. SHIM		AR
8	69-51845-3		. ACTUATOR, SENSOR	A-K	1
8	69-51845-4		. ACTUATOR, SENSOR	N-X	1
9	BACN10JC4		. NUT		3
10	AN960PD416L		. WASHER		3
11	NAS1104-8		. BOLT		3
12	69-37844-5		. HINGE ASSY	IV	1
12	69-37844-3		. HINGE ASSY (SB 57-1051, 1114)	B-HJK O-UWX	1
12	69-37844-1		. HINGE ASSY (PRE SB 57-1215)	AN	1
12	69-37844-3		. HINGE ASSY (POST SB 57-1215)	AN	1
13	69-37844-6		. . HINGE FTG (USED ON 69-37844-5)		1
13	69-37844-4		. . HINGE FTG (USED ON 69-37844-3)		1
13	69-37844-2		. . HINGE FTG (USED ON 69-37844-1)		1
14	BLFR8-026		. . BEARING, V81376 (BOEING 10-60545-140S)		1
14	KSBN8-21		. . BEARING, V97613 (BOEING 10-60545-140S)		1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USE CODE	QTY PER ASSY
1104-					
14	SBS16ATC32-2		. . BEARING, V21335 (BOEING 10-60545-140S)		1
14	YTA145		. . BEARING, V77896 (BOEING 10-60545-140S)		1
14	03-729-0500		. . BEARING, V09455 (BOEING 10-60545-140S)		1
15	BACS40G16A37		. SHIM		AR
16	BACS40G13A13		. SHIM		AR
17	BACN10JC4		. NUT		3
17	BACN10JC4		. NUT (USED WITH 69-37896-1,-3,-5,-7 ON 65-46424-29,-30)		4
18	NAS620-416		. WASHER		3
18	NAS679A4W		. WASHER (USED WITH 69-37896-1,-3,-5,-7 ON 65-46424-29,-30)		4
19	NAS1104-14		. BOLT		3
19	BACB30NF4-16		. BOLT (USED WITH 69-37896-1,-3,-5,-7 ON 65-46424-29,-30)		4
20	69-37896-5		. HINGE ASSY	IV	1
20	69-37896-7		. HINGE ASSY	KX	1
20	69-37896-3		. HINGE ASSY (SB 57-1051, 1114)	A-HJ	1
20	69-37896-1		. HINGE ASSY (OPT TO 69-37896-3)	AN	1
20	69-37896-1		. HINGE ASSY (PRE SB 57-1215)	AN	1
20	69-37896-3		. HINGE ASSY (POST SB 57-1215)	AN	1
21	69-37896-6		. . HINGE FTG (USED ON 69-37896-5)		1
21	69-37896-8		. . HINGE FTG (USED ON 69-37896-6)		1
21	69-37896-4		. . HINGE FTG (USED ON 69-37896-3)		1
21	69-37896-2		. . HINGE FTG (USED ON 69-37896-1)		1
22	BLFR8-026		. . BEARING, V81376 (BOEING 10-60545-140S)		1
22	KSBN8-21		. . BEARING, V97613 (BOEING 10-60545-140S)		1
22	SBS16ATC32-2		. . BEARING, V21335 (BOEING 10-60545-140S)		1
22	YTA145		. . BEARING, V77896 (BOEING 10-60545-140S)		1
22	03-729-0500		. . BEARING, V09455 (BOEING 10-60545-140S)		1
23	BACS40G13A38		. SHIM		AR
24	BACS40G13A13		. SHIM		AR

VENDORS

V09455	LEAR SIEGLER, INC., TRANSPORT DYNAMICS DIV., 3131 W. SEGERSTROM AVE., SANTA ANA, CALIFORNIA 92702
V21335	THE FAFNIR BEARING CO., DIVISION OF TEXTRON, INC., 37 BOOTH ST., NEW BRITAIN, CONNECTICUT 06050
V77896	REX CHAINBELT, INC., BEARING DIV., 2400 CURTIS ST., DOWNER'S GROVE, ILLINOIS 60515
V81376	SOUTHWEST PRODUCTS COMPANY, 1705 S. MOUNTAIN AVE., MONROVIA, CALIFORNIA 91016
V97613	SARGENT INDUSTRIES, KAHR BEARING DIV., 3010 N. SAN FERNANDO BLVD, BURBANK, CALIFORNIA 91503

Part No.	Fig. and Index No.	Qty. per Assy.
AN320-4		AR
AN960PD4161		AR
AN960PD5161		AR
BACB10A661	1102-31	2
BACB28X10C33	1101-16	1
BACB28X14B33	1101-17	1
BACB28X4B11	1103-17	2
BACB28X4B11	1103-20	2
BACB28X4B13	1103-7	1
BACB28X4B16	1103-8	1
BACB28X4B19	1101-18	4
BACB28X4B25	1102-32	4
BACB28Y7B16	1103-9	2
BACB30FM5-3	1101-25	8
BACB30FM6-3	1101-23	32
BACB301B6-5	1101-22	20
BACB301J4DU11	1103-14	3
BACB301J4DU16	1102-19	2
BACB301J4DU18	1102-4	2
BACB301J4DU18	1102-15	2
BACB301J4DU44	1102-28	2
BACB301J4D419	1102-54	1
BACB301U3-4	1103-31	5
BACB301U3-4	1101-2	6
BACB301U3-4	1101-2	8
BACB301U3-5	1101-7	14
BACB301U3-6	1101-7	14
BACB30M-5	1101-26	8
BACB30M-6	1101-24	32
BACB30NF4-14	1104-19	3
BACB30NF4-6	1103-4	2
BACB30NF4-7	1103-3	4
BACB30NF4-7	1103-23	1
BACB30NF4-8	1104-3	4
BACB30NF4-8	1104-11	3
BACB30NF4-9	1103-24	2
BACN10CP51	1102-10	1
BACN10JC3	1103-29	5
BACN10JC3	1101-1	6
BACN10JC3	1101-1	8
BACN10JC3	1101-6	14
BACN10JC4	1103-1	6
BACN10JC4	1103-21	3
BACN10JC4	1104-1	4
BACN10JC4	1104-9	3

Part No.	Fig. and Index No.	Qty. per Assy.
BACN10JC4	1104-17	3
BACN10JD104	1102-2	2
BACN10JD104	1102-13	2
BACN10JD104	1102-17	2
BACN10JD104	1102-26	2
BACN10JD104	1103-12	3
BACS40G13A13	1104-16	AR
BACS40G13A13	1104-24	AR
BACS40G13A38	1104-23	AR
BACS40G16A37	1104-15	AR
BACS40R10D10F	1104-7	AR
BACS40R10E23F	1103-28	AR
BACS40R10E26F	1103-10	AR
B1FR8-026	1104-14	1
B1FR8-026	1104-22	1
KSBN8-21	1104-14	1
KSBN8-21	1104-22	1
MS24665-134		AR
MS21232-4		AR
MS21232-6		AR
NAS1515M10		AR
NAS354-5-310		AR
NAS509-5		AR
NAS509-51		AR
NAS620-416		AR
NAS74A4-005P		AR
NAS77A4-11P		AR
SBS16ATC32-2	1104-14	1
SBS16ATC32-2	1104-22	1
YTA145	1104-14	1
YTA145	1104-22	1
YTA187	1102-8	1
03-729-0500	1104-14	1
03-729-0500	1104-22	1
10-60545-140S	1104-14	1
10-60545-140S	1104-22	1
10-60779-124	1102-8	1
10-60779-201	1102-24	2
10-60779-2011	1102-24	1

Part No.	Fig. and Index No.	Qty. per Assy.
65-46424-1	1101	
65-46424-10	1101	
65-46424-13	1101	
65-46424-14	1101	
65-46424-15	1101-19	1
65-46424-19	1101	
65-46424-2	1101	
65-46424-20	1101	
65-46424-21	1101	
65-46424-22	1101	
65-46424-23	1101	
65-46424-24	1101	
65-46424-25	1101	RF
65-46424-26	1101	RF
65-46424-27	1101	RF
65-46424-28	1101	RF
65-46424-29	1101	RF
65-46424-3	1101-19	1
65-46424-30	1101	RF
65-46424-5	1101	
65-46424-6	1101	
65-46424-7	1101	
65-46424-8	1101	
65-46424-9	1101	
65-49528-11	1102-29	2
65-49528-12	1102-30	1
65-49528-7	1102-29	2
65-49528-8	1102-30	1
65-49532-10	1101-14	1
65-49532-11	1101-15	1
65-49532-12	1101-15	1
65-49532-17	1101-14	1
65-49532-18	1101-14	1
65-49532-19	1101-15	1
65-49532-20	1101-15	1
65-49532-21	1101-14	1
65-49532-22	1101-14	1
65-49532-23	1101-14	1
65-49532-24	1101-14	1
65-49532-25	1101-15	1
65-49532-26	1101-15	1
65-49532-27	1101-14	1
65-49532-28	1101-14	1
65-49532-29	1101-15	1
65-49532-3	1101-15	1
65-49532-30	1101-15	1
65-49532-31	1101-14	1
65-49532-32	1101-14	1
65-49532-33	1101-15	1

Part No.	Fig. and Index No.	Qty. per Assy.
65-49532-34	1101-15	1
65-49532-4	1101-15	1
65-49532-5	1101-14	1
65-49532-6	1101-14	1
65-49532-7	1101-15	1
65-49532-8	1101-15	1
65-49532-9	1101-14	1
65-67101-3	1101-0	1
65-67101-4	1101-0	1
65-67101-5	1101-4	1
65-67101-6	1101-4	1
65-67101-23	1101-0	1
65-67101-24	1101-0	1
65-67101-29	1101-0	1
65-67101-29	1103-28A	1
65-67101-30	1101-0	1
65-67101-30	1103-28A	1
65-86048-1	1101-20	2
65-86048-1	1101-21	2
65-86048-2	1101-20	2
65-86048-2	1101-21	2
69-37244-3	1102-21	2
69-37808-1	1103-5	1
69-37808-10	1103-5	1
69-37808-11	1103-6	1
69-37808-12	1103-6	1
69-37808-13	1103-5	1
69-37808-14	1103-5	1
69-37808-15	1103-6	1
69-37808-16	1103-6	1
69-37808-2	1103-5	1
69-37808-3	1103-6	1
69-37808-4	1103-6	1
69-37808-9	1103-5	1
69-37841-11	1102-5	1
69-37841-13	1102-6	1
69-37841-7	1102-5	2
69-37841-9	1102-6	1
69-37842-1	1103-15	1
69-37842-12	1103-26	1
69-37842-13	1103-25	3
69-37842-14	1103-26	1
69-37842-2	1103-18	2
69-37842-6	1103-25	3
69-37842-7	1103-16	1
69-37842-8	1103-19	1
69-37843-1	1104-4	2
69-37843-2	1104-5	1
69-37843-3	1104-4	2

Part No.	Fig. and Index No.	Qty. per Assy.
69-37843-4	1104-5	1
69-37843-5	1104-4	2
69-37843-6	1104-5	1
69-37844-1	1104-12	1
69-37844-2	1104-13	1
69-37844-3	1104-12	1
69-37844-4	1104-13	1
69-37844-5	1104-12	1
69-37844-6	1104-13	1
69-37896-1	1104-20	1
69-37896-2	1104-21	1
69-37896-3	1104-20	1
69-37896-4	1104-21	1
69-37896-5	1104-20	1
69-37896-6	1104-21	1
69-37896-7	1104-20	1
69-37896-8	1104-21	1
69-43512-13	1101-13	1
69-43512-14	1101-13	1
69-43513-14	1101-8	1
69-43513-15	1101-9	2
69-43513-16	1101-10	1
69-43513-17	1101-11	1
69-51845-3	1104-8	1
69-51845-4	1104-8	1
69-54935-3	1101-5	1
69-54935-4	1101-5	1
69-54935-5	1101-5	1
69-54935-6	1101-5	1
69-54940-1	1101-3	1
69-54942-1	1101-14	1
69-54942-2	1101-14	1
69-54943-11	1101-12	1
69-54943-12	1101-12	1
69-54943-23	1101-12	1
69-54943-24	1101-12	1
69-54943-29	1101-13	1
69-54943-3	1101-12	1
69-54943-30	1101-13	1
69-54943-4	1101-12	1
69-54943-7	1101-12	1
69-54943-8	1101-12	1
69-55450-1	1102-58	1
69-61262-3	1102-21	2
69-61262-8	1102-22	1

Part No.	Fig. and Index No.	Qty. per Assy.
69-62782-1	1101-5	1
69-62782-2	1101-5	
69-62783-1	1101-4	1
69-62783-2	1101-4	1
69-62784-1	1101-3	1
69-62796-1	1103-32	1